Standard Aircraft Characteristics

NAVY MODEL UH-1N AIRCRAFT

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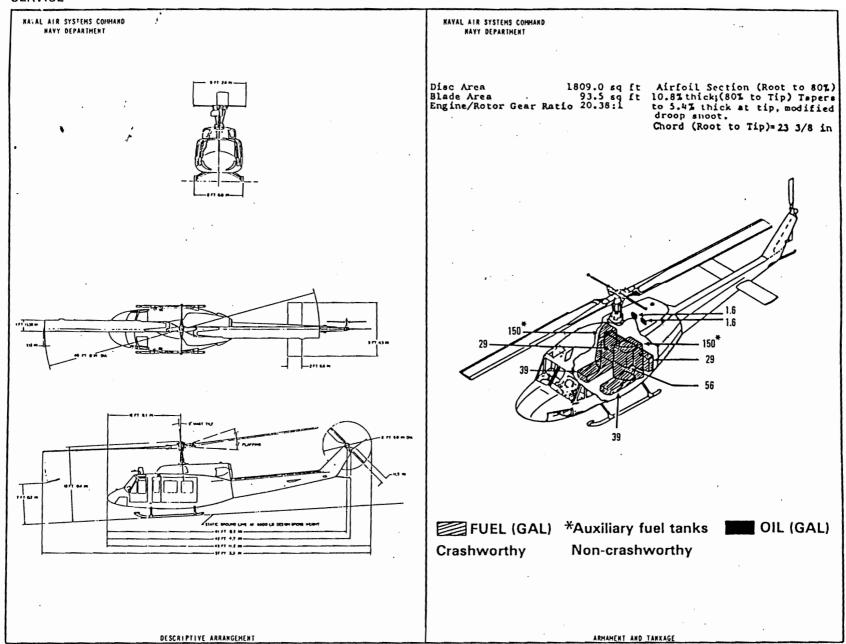
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STANDARD AIRCRAFT CHARACTERISTICS UH-1N BELL

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POWER PLANT	MISSION AND DESCRIPTION	WEIGHTS		
No. & Model (1) T400-CP-400 Manufacturer United Aircraft of Canada (Pratt and Whitney) Engine Spec. No 712 C Type Twin Section Free Pow Turbine with Reduction Cearbox Gear Reduction Ratios Main Rotor 20.383:1 Tail Rotor 3.974:1 RATINGS		Loading Weight L.T.		
-SHP RPM ALT Intermediate 1800 6600 SL Maximum Continuoua 1530 6600 SL Single Power Section Intermediate 900 6600 SL	The twin power section installation improves both hot day and altitude performance. The semi-monocoque fuselage is of all metal construction as are	· · · · · · · · · · · · · · · · · · ·		
Maximum Continuous 765 6600 SL Transmission Limits	the tail rotor blades and the two main rotor blades. The large sliding door along each side allows rapid entry and exit and sim-	FUEL AND OIL		
Intermediate 1290 SHP Maximum Continuous 1134 SHP	plified straight-through loading from either side or both sides simultaneously. The knee-high cargo floor also contributes to	FUEL		
ELECTRONICS	loading ease.	Gal. No. of Tanks Locations		
VIIF-FM Radio Set AN/ARC-114 Altimeter Encoder AU-21/A UNF-AM Radio Set AN/ARC-116 Communication Sys C-6533/ARC Gyromag Compass Set AN/ASN-43() UNF Dir Find Sys AN/ARA-50 TACAN Navigation Set AN/ARN-105 Transponder Set AN/APX-72 HF Radio Set AN/APX-72 AN/APC-102 Auto Direc Finder AN/ARN-89 Radar Altimeter AN/APN-171(V) Transponder Test Set TS-1843A/APX PROVISIONS FOR Loudspesker System Kit Commun Security Set TSEC/KY-28 VNF-AM Radio Set AN/ARC-115 Hark XII Computer KIT-1A/TSEC	DEVELOPMENT Contract Placement September 1969 First Flight (Comma. 212) April 1969 (USAF UH-1N) March 1970 (USN UH-1N) January 1971 First Delivery March 1971	191.9* 300.0 2 Cabin Fuel GradeJP-4/JP-5/JP-8 Fuel SpecMIL-J-5624 *Crashworthy Fuel System OIL Engine (Cal.)3.2 SpecMIL-L-23699 ORDNANCE		
	DIMENSIONS	ACCOMMODATIONS		
	Rotor Diameter 48.0' Length Rotors Operating 57.3' Rotors Static 57.3 Fuselage 42.4' Span (Max Lateral) 9.2' Height 14.9' Tread 8.5' Ground Clearence 7.0' (Static, Against Stops)	Crew (Observation)		
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DEDECTION AND CONTRACT CONTRAC										
	PERFORMANCE SUMMARY									
TAKE-OFF LOADING	CONDITION		1 OBSERVATION	2 RESCUE	3 MEDICAL EVACUATION	4 TROOP TRANSPORT	5 EXTERNAL TRANSPORT	6 FERRY		
TAKE-OFF WEIGHT		fb.	8582	8740	8622	9982	10500	10000		
Fuel Internal/external (JP-5)		lb./lb.	1305/0	1305/0	1305/0	1305/0	1305/0	3010		
Payload OUT/ RETURN		lb.	0/0	0/200	0/1200	1800/0	2318/0	0/0		
Disc loading	lb.	/sq. ft.	4.74	4.83	4,77	5.52	5.80	5.53		
Vertical rate of climb at S.L. (/	·)	fpm.	1855	1720	1835	800	425	810		
Absolute hovering ceiling (OGE)	(A)	ft.	14000	13400	13900	9200	5200	9250		
Max. rate of climb at S.L.	(A)	fpm.	2450	2400	2440	1730	1600	1750		
Service ceiling	(A)	ft.	21600	21200	21500	18000	16900	18100		
Speed at S.L.	(B)	kn.	121	119	121	109	106	109		
Max speed/altitude	(B)	kn./ft.	125/3500	123/3500	125/3500	113/3000	111/3000	113/3000		
O.E.I. Service ceiling	(A)	ft.	14200	13800	14100	10500	9200	10600		
Min. speed (O.E.I.)	(A),(D)	kn.	14	15	14	24	26	24		
Max. speed (O.E.I.)	(A), (D), (E)	kn.	110	109	110	102	98	102		
Combat radius		n. mi.	81	72	85	82	70	-		
Mission time		hrs.	1.48	1.54	1.62	1.45	1.60	-		
Average cruising speed		kn.	124	. 124	120	123	98	-		
Cruising altitude		ft.	SL	SL	5000	SL	SL	-		
Range	:	n. mi.	172	171	190	165	131	491		
Average cruising speed		kn,	124	124	121	115	80	118		
Cruising altitude		ft.	SL	\$L	5000	SL_	SL	2000 to 10000		
Maximum andurance	(c)	hrs.	2.00	2.00	2.25	1.93	1.74	5.87		
Endurance speed		kn.	56	56	57	59	50	58		

NOTES

5000

SL

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Endurance altitude

(A) At intermediate power or 1290 hp take-off transmission limit.
(B) At V_{NE}.
(C) Endurance time excludes engine atart and take-off, climb to altitude, and reserve.
(D) At sea level.
(E) At maximum continuous power.
(F) All performance st atandard day conditions.
(G) Engine specification fuel flow increased 5%.
(H) Data basis: UH-IN and Hodel 212 flight test.

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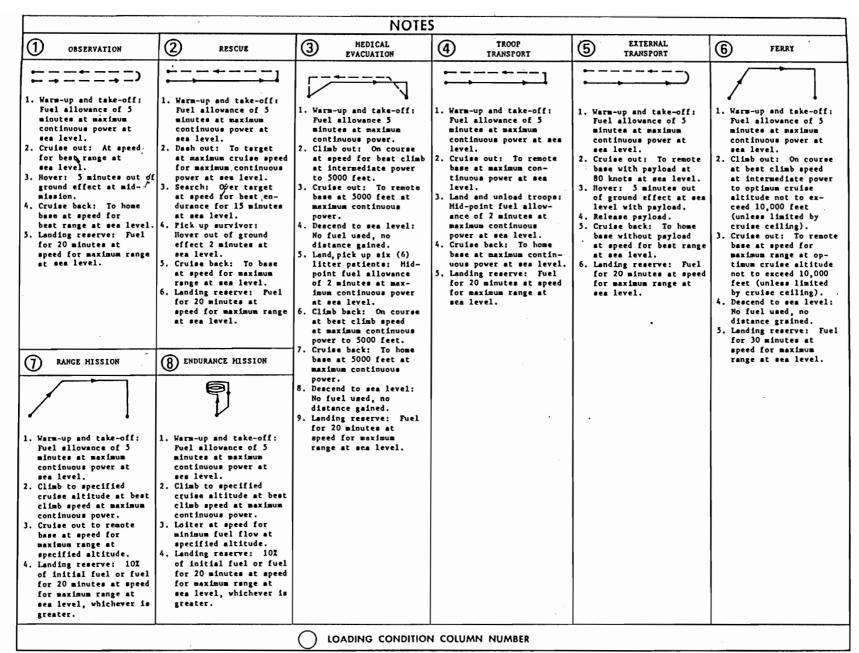
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2000 to 10000

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